The effect of blade/wall relative motion was not covered in Refs 5–8, but was investigated by Gearhart¹² and Dean¹⁰ and although their results have certain qualitative differences, possibly due to geometric variations, the convection effect due to the wall movement, though not as powerful as the effect of the presence of the gap alone, was seen to be important.

No cascade experiment, however, is able to investigate radial effects, centrifugal force on the blade boundary layer and the consequence of radial variations in the blade flow due to design considerations, so it remains to examine data from real turbomachines to evaluate these.

Conclusions

Part 1 of this review has covered about 30 years of research into tip gap phenomena in cascades. A good body of detailed, but uncoordinated work has resulted for diffusing type cascades, while that for accelerating type cascades is both sparse and lacking in the same detail. A good physical understanding of the flow in a tip gap region and the flow effect on conditions across the cascade channel have resulted, but the models proposed, as well as having a degree of mutual contradiction, all fail to deal adequately with the distribution of the tip vortex within the gap. In addition, while some workers have used ingenious methods to simulate rotor tip/wall relative motion, they have not been able to represent all the effects of rotation and may not have given a realistic representation of the casing boundary layer in rotor relative coordinates. Nevertheless, the detail with which it has been possible to make measurements in the comparatively easy conditions of a cascade has led to an understanding of the flow structure.

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References

- 1. **Peacock R. E.** A review of turbomachinery tip gap effects. Part 2: rotating machinery. *Int. J. Heat and Fluid Flow* (1983) 4(1), (in press)
- 2. Mikolajczak A. A. The practical importance of unsteady flow, Unsteady Phenomena in Turbomachinery. AGARD CP 177. September 1975
- 3. Williams D. D. and Yost J. Some aspects of inlet engine flow compatibility, J. Royal, Aero. Soc. LXXVII 1973
- Peacock R. E. and Overli J. Dynamic internal flows in compressors with pressure maldistributed inlet conditions. Unsteady Phenomena in Turbomachinery AGARD CP 177. 1975
- 5. Khabbaz G. R. The influence of tip clearance on stall limits of a rectilinear cascade of compressor blades. *M.I.T. Gas Turbine Laboratory Report No. 54. 1959*
- 6. Yokoyama E. Comparative study of tip clearance effects in compressors and turbines. M.I.T. Gas Turbine Laboratory Report No. 63. 1961
- 7. Lakshminarayana B. and Horlock J. H. Tip-clearance flow and losses for an isolated compressor blade. ARC R & M 3316. 1963
- 8. Lakshminarayana B. and Horlock J. H. Leakage and secondary flows in compressor cascades. ARC R & M 3483. 1967
- 9. Holme O. A. M. Measurements of the pressure distribution on rectangular wings of different aspect ratio. *Flygtecniska Försökanstalten FFA No. 37. 1950*
- 10. Dean R. C. Jr The influence of tip clearance on boundary layer flow in a rectilinear cascade. M.I.T. Gas Turbine Laboratory Report No. 27-3. 1954
- 11. Peacock R. E. Boundary layer suction to eliminate corner separation in cascades of aerofoils. ARC R & M 3663. 1971
- 12. Gearhart W. S. Tip clearance flow in turbomachines. The Pennsylvania State University Science and Engineering Ordnance Research Lab TM506.2491-04. 1964
- 13. Dring R. P., Joslyn H. D. and Hardin L. W. Compressor rotor aerodynamics—an analytical and experimental investigation. United Technologies Research Centre Report VTRC 80-15. 1980
- 14. Dring R. P., Joslyn H. D. and Hardin L. W. An investigation of axial compressor rotor aerodynamics. *The ASME Gas Turbine Conference, Paper No. ASME 81-GT-56. 1981*
- 15. Booth T. C., Dodge P. R. and Hepworth H. K. Rotor-tip leakage: Part I—Basic methodology. The ASME Gas Turbine Conference Paper No. 81-GT-71. 1981
- Wadia, A. R. and Booth T. C. Rotor-tip leakage: Part II— Design optimisation through viscous analysis and experiment. The ASME Gas Turbine Conference Paper 81-GT-72. 1981
- 17. Peacock R. E. Unpublished data measured at University of Cambridge, 1965
- Hawthorne W. R. Rotational flow through cascades. Part 1: The components of vorticity. The Quarterly Journal of Mechanics and Applied Mathematics, Vol VIII, Part 3, Sept. 1955

Corrigendum

In the Technical Note by R. Peretz titled 'Relation between evaporator and condenser lengths of a finless heat pipe to achieve a maximum heat flow per unit weight' published in the September issue (Volume 3, No 3), Eq. (14) contained $\frac{1}{2}\lambda$ rather than $(1/2\lambda)$ in both numberator and denominator. Thus the equation should have read:

 $\left(L_{\rm c}/L_{\rm c}\right)^2$

$$=\frac{h_{\rm o,c}d_{\rm 0})^{-1} + (1/2\lambda)\ln(d_{\rm o}/d_{\rm i}) + (h_{\rm i,c}d_{\rm i})^{-1}}{h_{\rm o,e}d_{\rm o})^{-1} + (1/2\lambda)\ln(d_{\rm o}/d_{\rm i}) + (h_{\rm i,e}d_{\rm i})^{-1}} \qquad (14)$$